

# Continuous Systems, Infinite Degrees of Freedom

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June 3, 2010

## Outline

### Continous Systems

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## Intro

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### Discrete models

Until now, structures were discretized, maybe lumping their masses in the *dynamical degrees of freedom* or maybe to use the *FEM* to derive a stiffness matrix, to be subjected to static condensation in the occurrence of lumped masses or, on the contrary, to be used *as is*.

Multistorey buildings are excellent examples of structures for which a few dynamical degrees of freedom can describe the dynamical response.

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### Continuous models

For different type of structures (e.g., bridges, chimneys), a lumped mass model is not the first option. While a *FE* model is always appropriate, there is no apparent way of lumping the structural masses in a way that is obviously correct, and a great number of degrees of freedom must be retained in the dynamic analysis.

An alternative to detailed *FE* models is deriving the equation of motion, in terms of partial derivatives differential equation, for the continuous systems.

# Continuous Systems

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There are many different continuous systems whose dynamics are approachable with the instruments of classical mechanics,

- ▶ taught strings,
- ▶ axially loaded rods,
- ▶ beams in flexure,
- ▶ plates and shells,
- ▶ 3D solids.

In the following, we will focus our interest on beams in flexure.

## EoM for an undamped beam

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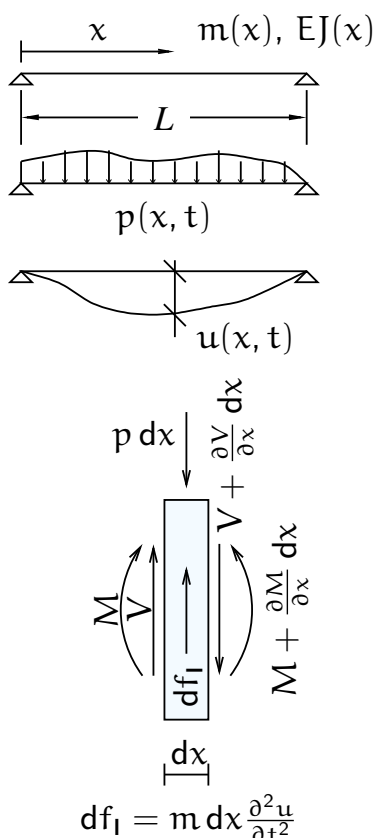
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At the left, a straight beam with characteristic depending on position  $x$ :  $m = m(x)$  and  $EJ = EJ(x)$ ; with the signs conventions for displacements, accelerations, forces and bending moments reported left, the equation of vertical equilibrium for an infinitesimal slice of beam is

$$V - (V + \frac{\partial V}{\partial x} dx) + m dx \frac{\partial^2 u}{\partial t^2} - p(x, t) dx = 0.$$

Rearranging and simplifying  $dx$ ,

$$\frac{\partial V}{\partial x} = m \frac{\partial^2 u}{\partial t^2} - p(x, t).$$

The rotational equilibrium, neglecting rotational inertia and simplifying  $dx$  is

$$\frac{\partial M}{\partial x} = V.$$

## Equation of motion, 2

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Deriving with respect to  $x$  both members of the rotational equilibrium equation, it is

$$\frac{\partial V}{\partial x} = \frac{\partial^2 M}{\partial x^2}$$

Substituting in the equation of vertical equilibrium and rearranging

$$m(x) \frac{\partial^2 u}{\partial t^2} - \frac{\partial^2 M}{\partial x^2} = p(x, t)$$

Using the moment-curvature relationship,

$$M = -EJ \frac{\partial^2 u}{\partial x^2}$$

and substituting in the equation above we have the equation of dynamic equilibrium

$$m(x) \frac{\partial^2 u}{\partial t^2} + \frac{\partial^2}{\partial x^2} \left[ EJ(x) \frac{\partial^2 u}{\partial x^2} \right] = p(x, t).$$

## Effective Earthquake Loading

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If our continuous structure is subjected to earthquake excitation, we will write, as usual,  $u_{TOT} = u(x, t) + u_g(t)$  and, consequently,

$$\ddot{u}_{TOT} = \ddot{u}(x, t) + \ddot{u}_g(t)$$

and, using the usual considerations,

$$p_{\text{eff}}(x, t) = -m(x) \ddot{u}_g(t).$$

In  $p_{\text{eff}}$  we have a separation of variables: in the case of earthquake excitation all the considerations we have done on expressing the response in terms of static modal responses and pseudo/acceleration response will be applicable.

Only a word of caution, in every case we must consider the component of earthquake acceleration *parallel* to the transverse motion of the beam.

## Free Vibrations

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For free vibrations,  $p(x, t) \equiv 0$  and the equation of equilibrium is

$$m(x) \frac{\partial^2 u}{\partial t^2} + \frac{\partial^2}{\partial x^2} \left[ EJ(x) \frac{\partial^2 u}{\partial x^2} \right] = 0.$$

Using separation of variables, with the following notations,

$$u(x, t) = q(t)\phi(x), \quad \frac{\partial u}{\partial t} = \dot{q}\phi, \quad \frac{\partial u}{\partial x} = q\phi'$$

etc for higher order derivatives, we have

$$m(x)\ddot{q}(t)\phi(x) + q(t) [EJ(x)\phi''(x)]'' = 0.$$

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Dividing both terms in

$$m(x)\ddot{q}(t)\phi(x) + q(t) [EJ(x)\phi''(x)]'' = 0.$$

by  $m(x)u(x, t) = m(x)q(t)\phi(x)$  and rearranging, we have

$$-\frac{\ddot{q}(t)}{q(t)} = \frac{[EJ(x)\phi''(x)]''}{m(x)\phi(x)}.$$

The left member is a function of time only, the right member a function of position only, and they are equal... this is possible if and only if both terms are constant, let's name this constant  $\omega^2$  and write

$$-\frac{\ddot{q}(t)}{q(t)} = \frac{[EJ(x)\phi''(x)]''}{m(x)\phi(x)} = \omega^2,$$

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Example

From the previous equations we can derive the following two

$$\ddot{q} + \omega^2 q = 0$$
$$[EJ(x)\phi''(x)]'' = \omega^2 m(x)\phi(x)$$

From the first,  $\ddot{q} + \omega^2 q = 0$ , it is apparent that free vibration shapes  $\phi(x)$  will be modulated by a trig function

$$q(t) = A \sin \omega t + B \cos \omega t.$$

To find something about  $\omega$ 's and  $\phi$ 's (that is, the eigenvalues and the *eigenfunctions* of our problem), we have to introduce an important simplification.

## Eigenpairs of a uniform beam

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Example

With  $EJ = \text{const.}$  and  $m = \text{const.}$ , we have from the second equation in previous slide,

$$EJ\phi^{IV} - \omega^2 m\phi = 0,$$

with  $\beta^4 = \frac{\omega^2 m}{EJ}$  it is

$$\phi^{IV} - \beta^4 \phi = 0$$

a differential equation of 4<sup>th</sup> order with constant coefficients.

Substituting  $\phi = \exp st$  and simplifying,

$$s^4 - \beta^4 = 0,$$

the roots of the associated polynomial are

$$s_1 = \beta, s_2 = -\beta, s_3 = i\beta, s_4 = -i\beta$$

and the general integral is

$$\phi(x) = A \sin \beta x + B \cos \beta x + C \sinh \beta x + D \cosh \beta x$$

# Constants of Integration

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For a uniform beam in free vibration, the general integral is

$$\phi(x) = \mathcal{A} \sin \beta x + \mathcal{B} \cos \beta x + \mathcal{C} \sinh \beta x + \mathcal{D} \cosh \beta x$$

In this expression we see 5 parameters, the 4 constants of integration and the wave number  $\beta$  (further consideration shows that the constants can be arbitrarily scaled).

In general for the transverse motion of a segment of beam supported at the extremes we can write exactly 4 equations expressing boundary conditions, either from kinematic or static considerations.

All these boundary conditions

- ▶ lead to linear, homogeneous equation where
- ▶ the coefficients of the equations depend on the parameter  $\beta$ .

# Eigenvalues and eigenfunctions

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Imposing the boundary conditions give a homogeneous linear system with coefficients depending on  $\beta$ , hence:

- ▶ a non trivial solution is possible only for particular values of  $\beta$ , for which the determinant of the matrix of coefficients is equal to zero and
- ▶ the constants are known within a proportionality factor.

In the case of *MDOF* systems, the determinantal equation is an algebraic equation of order  $N$ , giving exactly  $N$  eigenvalues, now the equation to be solved is a transcendental equation (examples from the next slide), with an infinity of solutions.

## Simply supported beam

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Example

Consider a simply supported uniform beam of length  $L$ , displacements at both ends must be zero, as well as the bending moments:

$$\begin{aligned}\phi(0) = \mathcal{B} + \mathcal{D} &= 0, & \phi(L) &= 0, \\ -EJ\phi''(0) = \beta^2 EJ(\mathcal{B} - \mathcal{D}) &= 0, & -EJ\phi''(L) &= 0.\end{aligned}$$

The conditions for the left support require that  $\mathcal{B} = \mathcal{D} = 0$

Now, we can write the equations for the right support as

$$\begin{aligned}\phi(L) = \mathcal{A} \sin \beta L + \mathcal{C} \sinh \beta L &= 0 \\ -EJ\phi''(L) = \beta^2 EJ(\mathcal{A} \sin \beta L - \mathcal{C} \sinh \beta L) &= 0\end{aligned}$$

or

$$\begin{bmatrix} +\sin \beta L & +\sinh \beta L \\ +\sin \beta L & -\sinh \beta L \end{bmatrix} \begin{Bmatrix} \mathcal{A} \\ \mathcal{C} \end{Bmatrix} = \begin{Bmatrix} 0 \\ 0 \end{Bmatrix}.$$

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Example

For the simply supported beam we have

$$\begin{bmatrix} +\sin \beta L & +\sinh \beta L \\ +\sin \beta L & -\sinh \beta L \end{bmatrix} \begin{Bmatrix} \mathcal{A} \\ \mathcal{C} \end{Bmatrix} = \begin{Bmatrix} 0 \\ 0 \end{Bmatrix}.$$

The determinant is  $-2 \sin \beta L \sinh \beta L$ , equating to zero with the understanding that  $\sinh \beta L \neq 0$  if  $\beta \neq 0$  results in

$$\sin \beta L = 0.$$

All positive  $\beta$  solutions are given by

$$\beta L = n\pi$$

with  $n = 1, \dots, \infty$ . We have an infinity of eigenvalues,

$$\beta_n = \frac{n\pi}{L} \text{ and } \omega_n = \beta^2 \sqrt{\frac{EJ}{m}} = n^2 \pi^2 \sqrt{\frac{EJ}{mL^4}}$$

and of eigenfunctions

$$\phi_1 = \sin \frac{\pi x}{L}, \phi_2 = \sin \frac{2\pi x}{L}, \phi_3 = \sin \frac{3\pi x}{L}, \dots$$



## Cantilever beam

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For  $x = 0$ , we have zero displacement and zero rotation

$$\phi(0) = \mathcal{B} + \mathcal{D} = 0, \quad \phi'(0) = \beta(\mathcal{A} + \mathcal{C}) = 0,$$

for  $x = L$ , both bending moment and shear must be zero

$$-EJ\phi''(L) = 0, \quad -EJ\phi'''(L) = 0.$$

Substituting the expression of the general integral, with  $\mathcal{D} = -\mathcal{B}$ ,  $\mathcal{C} = -\mathcal{A}$  from the left end equations, in the right end equations and simplifying

$$\begin{bmatrix} \sinh \beta L + \sin \beta L & \cosh \beta L + \cos \beta L \\ \cosh \beta L + \cos \beta L & \sinh \beta L - \sin \beta L \end{bmatrix} \begin{Bmatrix} \mathcal{A} \\ \mathcal{B} \end{Bmatrix} = \begin{Bmatrix} 0 \\ 0 \end{Bmatrix}.$$

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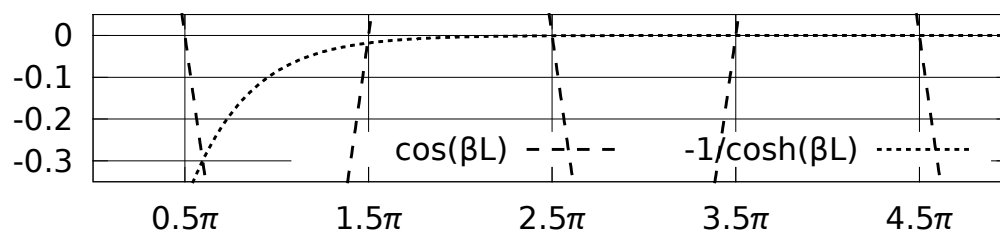
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Example

Imposing a zero determinant results in

$$\begin{aligned} (\cosh^2 \beta L - \sinh^2 \beta L) + (\sin^2 \beta L + \cos^2 \beta L) + 2 \cos \beta L \cosh \beta L &= \\ = 2(1 + \cos \beta L \cosh \beta L) &= 0 \end{aligned}$$

Rearranging,  $\cos \beta L = -(\cosh \beta L)^{-1}$  and plotting these functions on the same graph



it is  $\beta_1 L = 1.8751$  and  $\beta_2 L = 4.6941$ , while for  $n = 3, 4, \dots$  with good approximation it is  $\beta_n L \approx \frac{2n-1}{2}\pi$ .

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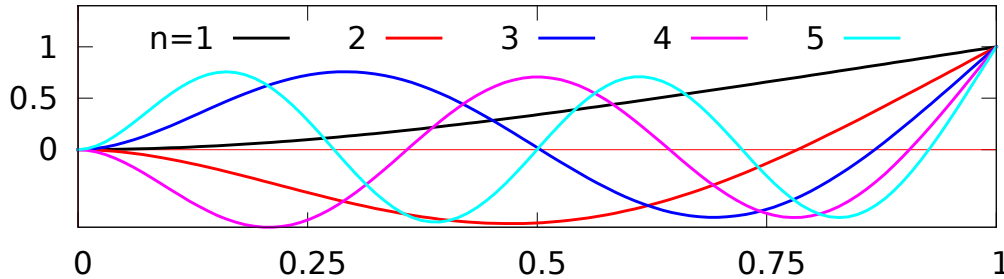
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Example

Eigenvectors are given by

$$\phi_n(x) = C_n \left[ (\cosh \beta_n x - \cos \beta_n x) - \frac{\cosh \beta_n L + \cos \beta_n L}{\sinh \beta_n L + \sin \beta_n L} (\sinh \beta_n x - \sin \beta_n x) \right]$$



Above, in abscissas  $x/L$  and in ordinates  $\phi_n(x)$  for the first 5 modes of vibration of the cantilever beam.

n	1	2	3	4	5
$\beta_n L$	1.8751	4.6941	7.8548	10.9962	$\approx 4.5\pi$
$\omega \sqrt{\frac{mL^4}{EJ}}$	3.516	22.031	61.70	120.9	...

# Mode Orthogonality

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Example

We will demonstrate mode orthogonality for a restricted set of boundary conditions, i.e., disregarding elastic supports and supported masses. In the beginning we have, for  $n = r$ ,

$$[EJ(x)\phi_r''(x)]'' = \omega_r^2 m(x)\phi_r(x)$$

premultiply both members by  $\phi_s(x)$  and integrating over the length of the beam gives

$$\int_0^L \phi_s(x) [EJ(x)\phi_r''(x)]'' dx = \omega_r^2 \int_0^L \phi_s(x) m(x)\phi_r(x) dx$$

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The left member can be integrated by parts, two times, as in

$$\int_0^L \phi_s(x) [EJ(x)\phi_r''(x)]'' dx =$$
$$\left[ \phi_s(x) [EJ(x)\phi_r''(x)]' \right]_0^L - \left[ \phi_s'(x) EJ(x)\phi_r''(x) \right]_0^L +$$
$$\int_0^L \phi_s''(x) EJ(x)\phi_r''(x) dx$$

but the terms in brackets are always zero, the first being the product of end displacement by end shear, the second the product of end rotation by bending moment, and for fixed constraints or free end one of the two terms must be zero. So it is

$$\int_0^L \phi_s''(x) EJ(x)\phi_r''(x) dx = \omega_r^2 \int_0^L \phi_s(x) m(x)\phi_r(x) dx.$$

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We write the last equation exchanging the roles of r and s and subtract from the original,

$$\int_0^L \phi_s''(x) EJ(x)\phi_r''(x) dx - \int_0^L \phi_r''(x) EJ(x)\phi_s''(x) dx =$$
$$\omega_r^2 \int_0^L \phi_s(x) m(x)\phi_r(x) dx - \omega_s^2 \int_0^L \phi_r(x) m(x)\phi_s(x) dx.$$

This obviously can be simplified giving

$$(\omega_r^2 - \omega_s^2) \int_0^L \phi_r(x) m(x)\phi_s(x) dx = 0$$

implying that, for  $\omega_r^2 \neq \omega_s^2$  the modes are orthogonal with respect to the mass distribution and the bending stiffness distribution.

## Forced dynamic response

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With  $u(x, t) = \sum_1^{\infty} \phi_m(x) q_m(t)$ , the equation of motion can be written

$$\sum_1^{\infty} m(x) \phi_m(x) \ddot{q}_m(t) + \sum_1^{\infty} [EJ(x) \phi_m''(x)]'' q_m(t) = p(x, t)$$

premultiplying by  $\phi_n$  and integrating each sum and the loading term

$$\sum_1^{\infty} \int_0^L \phi_n(x) m(x) \phi_m(x) \ddot{q}_m(t) dx + \sum_1^{\infty} \int_0^L \phi_n(x) [EJ(x) \phi_m''(x)]'' q_m(t) dx = \int_0^L \phi_n(x) p(x, t) dx$$

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Example

By the orthogonality of the eigenfunctions this can be simplified to

$$m_n \ddot{q}_n(t) + k_n q_n(t) = p_n(t), \quad n = 1, 2, \dots, \infty$$

with

$$m_n = \int_0^L \phi_n m \phi_n dx, \quad k_n = \int_0^L \phi_n [EJ \phi_n'']'' dx,$$

$$\text{and} \quad p_n(t) = \int_0^L \phi_n p(x, t) dx.$$

For free ends and/or fixed supports,  $k_n = \int_0^L \phi_n'' EJ \phi_n'' dx$ .

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Consider an effective earthquake load,  $p(x, t) = m(x)\ddot{u}_g(t)$ , with

$$\mathcal{L}_n = \int_0^L \phi_n(x) m(x) dx, \quad \Gamma_n = \frac{\mathcal{L}_n}{m_n},$$

the modal equation of motion can be written (with an obvious generalisation)

$$\ddot{q}_n + 2\omega_n \zeta_n \dot{q}_n + \omega_n^2 q_n = -\Gamma_n \ddot{u}_g(t)$$

and the modal response can be written, also for the case of continuous structures, as the product of the modal participation factor and the deformation response,

$$q_n(t) = \Gamma_n D_n(t).$$

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Modal contributions can be computed directly, e.g.

$$u_n(x, t) = \Gamma_n \phi_n(x) D_n(t),$$
$$M_n(x, t) = -\Gamma_n EJ(x) \phi_n''(x) D_n(t),$$

or can be computed from the equivalent static forces,

$$f_s(x, t) = [EJ(x)u(x, t)''']''.$$

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The modal contributions to equiv. static forces are

$$f_{sn}(x, t) = \Gamma_n [EJ(x)\phi_n(x)''']'' D_n(t),$$

that, because it is

$$[EJ(x)\phi''(x)]'' = \omega^2 m(x)\phi(x)$$

can be written in terms of the mass distribution and of the pseudo-acceleration response  $A_n(t) = \omega_n^2 D_n(t)$

$$f_{sn}(x, t) = \Gamma_n m(x)\phi_n(x)\omega_n^2 D_n(t) = \Gamma_n m(x)\phi_n(x)A_n(t).$$

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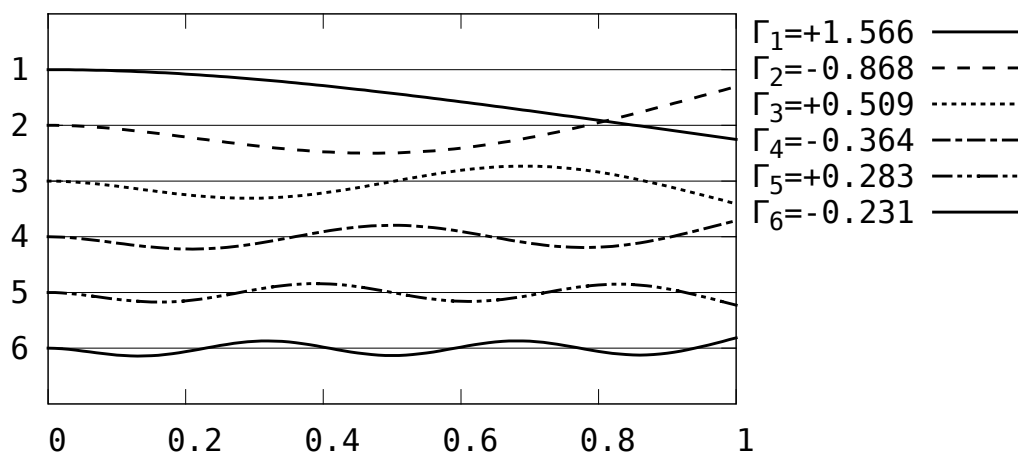
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Example

The effective load is proportional to the mass distribution, and we can do a modal mass decomposition in the same way that we had for *MDOF* systems,

$$m(x) = \sum r_n(x) = \sum \Gamma_n m(x)\phi_n(x)$$



Above, the modal mass decomposition  $r_n = \Gamma_n m\phi_n$ , for the first six modes of a uniform cantilever, in abscissa  $x/L$ .

## EQ example, cantilever

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Example

For a cantilever, it is possible to derive explicitly some response quantities,

$$V(x), \quad V_B, \quad M(x), \quad M_B,$$

that is, the shear force and the base shear force, the bending moment and the base bending moment.

$$V_n^{\text{st}}(x) = \int_x^L r_n(s) ds, \quad V_B^{\text{st}} = \int_0^L r_n(s) ds = \Gamma_n \mathcal{L}_n = M_n^*,$$

$$M_n^{\text{st}}(x) = \int_x^L r_n(s)(s-x) ds, \quad M_B^{\text{st}} = \int_0^L s r_n(s) ds = M_n^* h_n^*.$$

$M_n^*$  is the *participating modal mass* and expresses the participation of the different modes to the base shear, it is  $\sum M_n^* = \int_0^L m(x) dx$ .

$M_n^* h_n^*$  expresses the modal participation to base moment,  $h_n^*$  is the height where the participating modal mass  $M_n^*$  must be placed so that its effects on the base are the same of the static modal forces effects, or  $M_n^*$  is the resultant of s.m.f. and  $h_n^*$  is the position of this resultant.

## EQ example, cantilever, 2

Continuous Systems, Infinite Degrees of Freedom

Giacomo Boffi

Continuous Systems

Beams in Flexure

Equation of motion  
Earthquake Loading  
Free Vibrations  
Eigenpairs of a Uniform Beam  
Simply Supported Beam  
Cantilever Beam  
Mode Orthogonality  
Forced Response  
Earthquake Response

Example

Starting with the definition of total mass and operating a chain of substitutions,

$$M_{\text{TOT}} = \int_0^L m(x) dx = \sum \int_0^L r_n(x) dx$$

$$= \sum \int_0^L \Gamma_n m(x) \phi_n(x) dx = \sum \Gamma_n \int_0^L m(x) \phi_n(x) dx$$

$$= \sum \Gamma_n \mathcal{L}_n = \sum M_n^*,$$

we have demonstrated that the sum of the participating modal mass is equal to the total mass.

The demonstration that  $M_{B,\text{TOT}} = \sum M_n^* h_n^*$  is similar and is left as an exercise.

## EQ example, cantilever, 3

Continuous  
Systems, Infinite  
Degrees of  
Freedom

Giacomo Boffi

Continous  
Systems

Beams in Flexure

Equation of  
motion

Earthquake  
Loading

Free Vibrations

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Simply Supported  
Beam

Cantilever Beam  
Mode

Orthogonality

Forced Response

Earthquake  
Response

Example

For the first 6 modes of a uniform cantilever,

$n$	$\mathcal{L}_n$	$m_n$	$\Gamma_n$	$V_{B,n}$	$h_n$	$M_{B,n}$
1	0.391496	0.250	1.565984	0.613076	0.726477	0.445386
2	-0.216968	0.250	-0.867872	0.188300	0.209171	0.039387
3	0.127213	0.250	0.508851	0.064732	0.127410	0.008248
4	-0.090949	0.250	-0.363796	0.033087	0.090943	0.003009
5	0.070735	0.250	0.282942	0.020014	0.070736	0.001416
6	-0.057875	0.250	-0.231498	0.013398	0.057875	0.000775
7	0.048971	0.250	0.195883	0.009593	0.048971	0.000470
8	-0.042441	0.250	-0.169765	0.007205	0.042442	0.000306

The convergence for MB is faster than for  $V_B$ , because the latter is proportional to a higher derivative of displacements.