Continuous Systems, Infinite Degrees of Freedom

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Outline

Intro

Discrete models

Until now we have described or approximated the structural behaviour using the *dynamical degrees of freedom*, either directly contructing a model with lumped masses or using the *FEM* to derive a stiffness matrix and a *consistent mass matrix* or using the *FEM* stiffness with a lumped mass matrix reducing the degrees of freedom with the procedure of static condensation.

Multistory buildings are ecellent examples of structures for which a few dynamical degrees of freedom can describe the dynamical response, using only 3 deegres of freedom for each storey under the assumption of fully rigid floors.

Intro

Continuous models

For some type of structures (e.g., bridges, chimneys) a lumped mass model is not the first option.
While a *FE* model is however appropriate, there is no apparent way of lumping the structural masses in a way that is obviously correct, and a great number of degrees of freeedom must be retained in the dynamic analysis.
An alternative to detailed *FE* models is deriving the equation of motion for the continuous systems in terms of partial derivatives differential equation.

Continuous Systems

The equation of motion can be written in terms of partial derivatives for many different types of continuous systems, e.g.,

- taught strings,
- axially loaded rods,
- beams in flexure,
- plates and shells,
- 3D solids.

Today we will focus our interest on beams in flexure.

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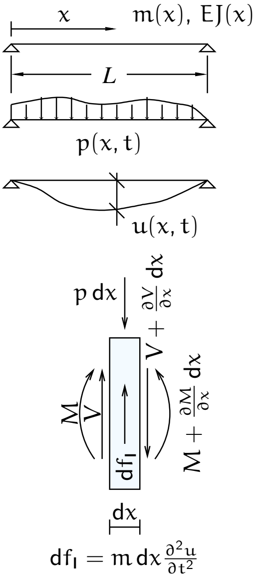
EoM for an undamped beam



At the left, a straight beam with characteristic depending on position x: $m = m(x)$ and $EJ = EJ(x)$; with the signs conventions for displacements, accelerations, forces and bending moments reported left, the equation of vertical equilibrium for an infinitesimal slice of beam is

$$V - (V + \frac{\partial V}{\partial x}\,dx) + m\,dx\,\frac{\partial^2 u}{\partial t^2} - p(x,t)\,dx = 0.$$

Rearranging and simplifying dx,

$$\frac{\partial V}{\partial x} = m\frac{\partial^2 u}{\partial t^2} - p(x,t).$$

The rotational equilibrium, neglecting rotational inertia and simplifying dx is

$$\frac{\partial M}{\partial x} = V.$$

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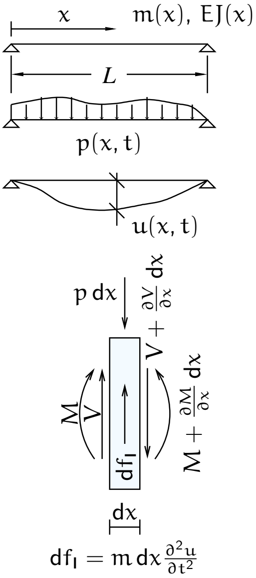

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$$\frac{\partial M}{\partial x} = V.$$

Equation of motion, 2

Deriving with respect to x both members of the rotational equilibrium equation, it is

$$\frac{\partial V}{\partial x} = \frac{\partial^2 M}{\partial x^2}$$

Substituting in the equation of vertical equilibrium and rearranging

$$m(x)\frac{\partial^2 u}{\partial t^2} - \frac{\partial^2 M}{\partial x^2} = p(x, t)$$

Using the moment-curvature relationship,

$$M = -EJ\frac{\partial^2 u}{\partial x^2}$$

and substituting in the equation above we have the equation of dynamic equilibrium

$$m(x)\frac{\partial^2 u}{\partial t^2} + \frac{\partial^2}{\partial x^2}\left[EJ(x)\frac{\partial^2 u}{\partial x^2}\right] = p(x, t).$$

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Effective Earthquake Loading

If our continuous structure is subjected to earthquake excitation, we will write, as usual, $u_{tot} = u(x,t) + u_g(t)$ and, consequently,

$$\ddot{u}_{tot} = \ddot{u}(x,t) + \ddot{u}_g(t)$$

and, using the usual considerations,

$$p_{eff}(x,t) = -m(x)\ddot{u}_g(t).$$

In p_{eff} we have a separation of variables: in the case of earthquake excitation all the considerations we have done on expressing the response in terms of static modal responses and pseudo/acceleration response will be applicable.
Only a word of caution, in every case we must consider the component of earthquake acceleration *parallel* to the transverse motion of the beam.

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Only a word of caution, in every case we must consider the component of earthquake acceleration *parallel* to the transverse motion of the beam.

Free Vibrations

For free vibrations, $p(x,t) \equiv 0$ and the equation of equilibrium is

$$m(x)\frac{\partial^2 u}{\partial t^2} + \frac{\partial^2}{\partial x^2}\left[EJ(x)\frac{\partial^2 u}{\partial x^2}\right] = 0.$$

Using separation of variables, with the following notations,

$$u(x,t) = q(t)\phi(x),\ \frac{\partial u}{\partial t} = \dot{q}\phi,\ \frac{\partial u}{\partial x} = q\phi'$$

etc for higher order derivatives, we have

$$m(x)\ddot{q}(t)\phi(x) + q(x)\left[EJ(x)\phi''\right]'' = 0.$$

Free Vibrations, 2

Dividing both terms in

$$m(x)\ddot{q}(t)\phi(x) + q(t)\left[EJ(x)\phi''(x)\right]'' = 0.$$

by $m(x)u(x,t) = m(x)q(t)\phi(x)$ and rearranging, we have

$$-\frac{\ddot{q}(t)}{q(t)} = \frac{[EJ(x)\phi''(x)]''}{m(x)\phi(x)}.$$

The left member is a function of time only, the right member a function of position only, and they are equal... this is possible if and only if both terms are constant, let's name this constant ω^2 and write

$$-\frac{\ddot{q}(t)}{q(t)} = \frac{[EJ(x)\phi''(x)]''}{m(x)\phi(x)} = \omega^2,$$

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Free Vibrations, 3

From the previous equations we can derive the following two

$$\ddot{q} + \omega^2 q = 0$$
$$\left[EJ(x)\phi''(x)\right]'' = \omega^2 m(x)\phi(x)$$

From the first, $\ddot{q} + \omega^2 q = 0$, it is apparent that free vibration shapes $\phi(x)$ will be modulated by a trig function

$$q(t) = A \sin \omega t + B \cos \omega t.$$

To find something about ω's and ϕ's (that is, the eigenvalues and the *eigenfunctions* of our problem), we have to introduce an important simplification.

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Eigenpairs of a uniform beam

With $EJ = \text{const.}$ and $m = \text{const.}$, we have from the second equation in previous slide,

$$EJ\phi^{IV} - \omega^2 m\phi = 0,$$

with $\beta^4 = \frac{\omega^2 m}{EJ}$ it is

$$\phi^{IV} - \beta^4\phi = 0$$

a differential equation of 4^{th} order with constant coefficients.
Substituting $\phi = \exp st$ and simplyfing,

$$s^4 - \beta^4 = 0,$$

the roots of the associated polynomial are

$$s_1 = \beta,\ s_2 = -\beta,\ s_3 = i\beta,\ s_4 = -i\beta$$

and the general integral is

$$\phi(x) = \mathcal{A}\sin\beta x + \mathcal{B}\cos\beta x + \mathcal{C}\sinh\beta x + \mathcal{D}\cosh\beta x$$

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Constants of Integration

For a uniform beam in free vibration, the general integral is

$$\phi(x) = \mathcal{A} \sin \beta x + \mathcal{B} \cos \beta x + \mathcal{C} \sinh \beta x + \mathcal{D} \cosh \beta x$$

In this expression we see 5 parameters, the 4 constants of integration and the wave number β (further consideration shows that the constants can be arbitrarily scaled).
In general for the transverse motion of a segment of beam supported at the extremes we can write exactly 4 equations expressing boundary conditions, either from kinematc or static considerations.
All these boundary conditions

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Eigenvalues and eigenfunctions

Imposing the boundary conditions give a homogeneous linear system with coefficients depending on β, hence:

- a non trivial solution is possible only for particular values of β, for which the determinant of the matrix of cofficients is equal to zero and
- the constants are known within a proportionality factor.

In the case of *MDOF* systems, the determinantal equation is an algebraic equation of order N, giving exactly N eigenvalues, now the equation to be solved is a trascendental equation (examples from the next slide), with an infinity of solutions.

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Simply supported beam

Consider a simply supported uniform beam of lenght L, displacements at both ends must be zero, as well as the bending moments:

$$\phi(0) = \mathcal{B} + \mathcal{D} = 0, \qquad \phi(L) = 0,$$
$$-EJ\phi''(0) = \beta^2 EJ(\mathcal{B} - \mathcal{D}) = 0, \qquad -EJ\phi''(L) = 0.$$

The conditions for the left support require that $\mathcal{B} = \mathcal{D} = 0$
Now, we can write the equations for the right support as

$$\phi(L) = \mathcal{A} \sin \beta L + \mathcal{C} \sinh \beta L = 0$$
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or

$$\begin{bmatrix} +\sin \beta L & +\sinh \beta L \\ +\sin \beta L & -\sinh \beta L \end{bmatrix} \begin{Bmatrix} \mathcal{A} \\ \mathcal{C} \end{Bmatrix} = \begin{Bmatrix} 0 \\ 0 \end{Bmatrix}.$$

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The determinant is $-2\sin\beta L \sinh\beta L$, equating to zero with the understanding that $\sinh\beta L \neq 0$ if $\beta \neq 0$ results in

$$\sin\beta L = 0.$$

All positive β solutions are given by

$$\beta L = n\pi$$

with $n = 1, \ldots, \infty$. We have an infinity of eigenvalues,

$$\beta_n = \frac{n\pi}{L} \text{ and } \omega_n = \beta^2\sqrt{\frac{EJ}{m}} = n^2\pi^2\sqrt{\frac{EJ}{mL^4}}$$

and of eigenfunctions

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Cantilever beam

For $x = 0$, we have zero displacement and zero rotation

$$\phi(0) = \mathcal{B} + \mathcal{D} = 0, \qquad \phi'(0) = \beta(\mathcal{A} + \mathcal{C}) = 0,$$

for $x = L$, both bending moment and shear must be zero

$$-EJ\phi''(L) = 0, \qquad -EJ\phi'''(L) = 0.$$

Substituting the expression of the general integral, with $\mathcal{D} = -\mathcal{B}$, $\mathcal{C} = -\mathcal{A}$ from the left end equations, in the right end equations and simplifying

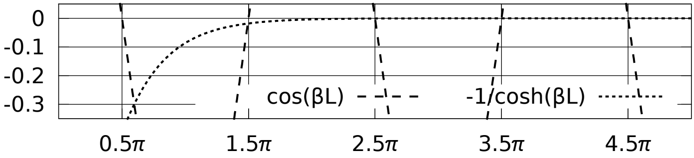
$$\begin{bmatrix} \sinh\beta L + \sin\beta L & \cosh\beta L + \cos\beta L \\ \cosh\beta L + \cos\beta L & \sinh\beta L - \sin\beta L \end{bmatrix} \begin{Bmatrix} \mathcal{A} \\ \mathcal{B} \end{Bmatrix} = \begin{Bmatrix} 0 \\ 0 \end{Bmatrix}.$$

Cantilever beam, 2

Imposing a zero determinant results in

$$(\cosh^2\beta L - \sinh^2\beta L) + (\sin^2\beta L + \cos^2\beta L) + 2\cos\beta L\cosh\beta L = \\ = 2(1+\cos\beta L\cosh\beta L) = 0$$

Rearranging, $\cos\beta L = -(\cosh\beta L)^{-1}$ and plotting these functions on the same graph



it is $\beta_1 L = 1.8751$ and $\beta_2 L = 4.6941$, while for $n = 3, 4, \ldots$ with good approximation it is $\beta_n L \approx \frac{2n-1}{2}\pi$.

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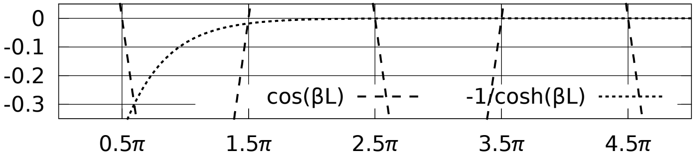

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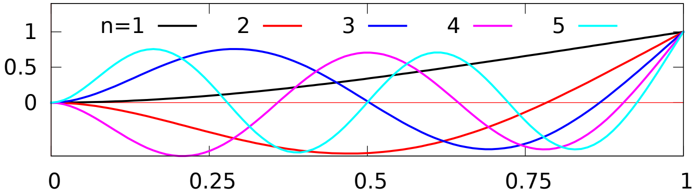
Cantilever beam, 3

Eigenvectors are given by

$$\phi_n(x) = C_n\left[(\cosh\beta_n x - \cos\beta_n x) - \frac{\cosh\beta_n L + \cos\beta_n L}{\sinh\beta_n L + \sin\beta_n L}(\sinh\beta_n x - \sin\beta_n x)\right]$$



Above, in abscissas x/L and in ordinates $\phi_n(x)$ for the first 5 modes of vibration of the cantilever beam.

n	1	2	3	4	5
$\beta_n L$	1.8751	4.6941	7.8548	10.9962	$\approx 4.5\pi$
$\omega\sqrt{\frac{mL^4}{EJ}}$	3.516	22.031	61.70	120.9	$\cdots$

Mode Orthogonality

We will demonstrate mode orhogonality for a restricted set of of boundary conditions, i.e., disregarding elastic supports and supported masses. In the beginning we have, for $n = r$,

$$\left[EJ(x)\phi_r''(x)\right]'' = \omega_r^2 m(x)\phi_r(x)$$

premultiply both members by $\phi_s(x)$ and integrating over the length of the beam gives

$$\int_0^L \phi_s(x)\left[EJ(x)\phi_r''(x)\right]'' dx = \omega_r^2 \int_0^L \phi_s(x)m(x)\phi_r(x)\, dx$$

Mode Orthogonality, 2

The left member can be integrated by parts, two times, as in

$$\int_0^L \phi_s(x) \left[EJ(x)\phi_r''(x)\right]'' dx =$$

$$\left[\phi_s(x) \left[EJ(x)\phi_r''(x)\right]'\right]_0^L - \left[\phi_s'(x)EJ(x)\phi_r''(x)\right]_0^L +$$

$$\int_0^L \phi_s''(x)EJ(x)\phi_r''(x)\, dx$$

but the terms in brackets are always zero, the first being the product of end displacement by end shear, the second the product of end rotation by bending moment, and for fixed constraints or free end one of the two terms must be zero. So it is

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Mode Orthogonality, 3

We write the last equation exchanging the roles of r and s and subtract from the original,

$$\int_0^L \phi_s''(x) EJ(x) \phi_r''(x)\, dx - \int_0^L \phi_r''(x) EJ(x) \phi_s''(x)\, dx =$$
$$\omega_r^2 \int_0^L \phi_s(x) m(x) \phi_r(x)\, dx - \omega_s^2 \int_0^L \phi_r(x) m(x) \phi_s(x)\, dx.$$

This obviously can be simplyfied giving

$$(\omega_r^2 - \omega_s^2) \int_0^L \phi_r(x) m(x) \phi_s(x)\, dx = 0$$

implying that, for $\omega_r^2 \neq \omega_s^2$ the modes are orthogonal with respect to the mass distribution and the bending stiffness distribution.

Forced dynamic response

With $u(x,t) = \sum_1^\infty \phi_m(x) q_m(t)$, the equation of motion can be written

$$\sum_1^\infty m(x)\phi_m(x)\ddot{q}_m(t) + \sum_1^\infty \left[EJ(x)\phi_m''(x)\right]'' q_m(t) = p(x,t)$$

premultiplying by ϕ_n and integrating each sum and the loading term

$$\sum_1^\infty \int_0^L \phi_n(x) m(x) \phi_m(x) \ddot{q}_m(t)\,dx +$$

$$\sum_1^\infty \int_0^L \phi_n(x) \left[EJ(x)\phi_m''(x)\right]'' q_m(t)\,dx = \int_0^L \phi_n(x) p(x,t)\,dx$$

Forced dynamic response, 2

By the orthogonality of the eigenfunctions this can be simplyfied to

$$m_n \ddot{q}_n(t) + k_n q_n(t) = p_n(t), \qquad n = 1, 2, \ldots, \infty$$

with

$$m_n = \int_0^L \phi_n m \phi_n \, dx, \qquad k_n = \int_0^L \phi_n \left[EJ\phi_n'' \right]'' dx,$$

and

$$p_n(t) = \int_0^L \phi_n p(x,t) \, dx.$$

For free ends and/or fixed supports, $k_n = \int_0^L \phi_n'' EJ \phi_n'' \, dx$.

Earthquake response

Consider an effective earthquake load, $p(x,t) = m(x)\ddot{u}_g(t)$, with

$$\mathcal{L}_n = \int_0^L \phi_n(x) m(x)\, dx, \qquad \Gamma_n = \frac{\mathcal{L}_n}{m_n},$$

the modal equation of motion can be written (with an obvious generalisation)

$$\ddot{q}_n + 2\omega_n\zeta_n\dot{q}_n + \omega_n^2 q = -\Gamma_n\ddot{u}_g(t)$$

and the modal response can be written, also for the case of continuous structures, as the product of the modal partecipation factor and the deformation response,

$$q_n(t) = \Gamma_n D_n(t).$$

Earthquake response, 2

Modal contributions can be computed directly, e.g.

$$u_n(x,t) = \Gamma_n \phi_n(x) D_n(t),$$
$$M_n(x,t) = -\Gamma_n EJ(x) \phi_n''(x) D_n(t),$$

or can be computed from the equivalent static forces,

$$f_s(x,t) = \left[EJ(x) u(x,t)''\right]''.$$

Earthquake response, 3

The modal contributions to equiv. static forces are

$$f_{sn}(x,t) = \Gamma_n \left[EJ(x)\phi_n(x)''\right]'' D_n(t),$$

that, because it is

$$\left[EJ(x)\phi''(x)\right]'' = \omega^2 m(x)\phi(x)$$

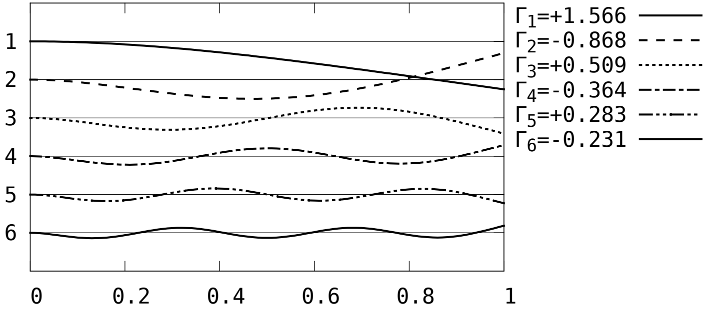
can be written in terms of the mass distribution and of the pseudo-acceleration response $A_n(t) = \omega_n^2 D_n(t)$

$$f_{sn}(x,t) = \Gamma_n m(x)\phi_n(x)\omega_n^2 D_n(t) = \Gamma_n m(x)\phi_n(x)A_n(t).$$

Earthquake response, 4

The effective load is proportional to the mass distribution, and we can do a modal mass decomposition in the same way that we had for *MDOF* systems,

$$m(x) = \sum r_n(x) = \sum \Gamma_n m(x) \phi_n(x)$$



Above, the modal mass decomposition $r_n = \Gamma_n m \phi_n$,for the first six modes of a uniform cantilever, in abscissa x/L.

EQ example, cantilever

For a cantilever, it is possible to derive explicitly some response quantities,

$$V(x), \qquad V_B, \qquad M(x), \qquad M_B,$$

that is, the shear force and the base shear force, the bending moment and the base bending moment.

$$V_n^{st}(x) = \int_x^L r_n(s)\,ds, \qquad V_B^{st} = \int_0^L r_n(s)\,ds = \Gamma_n \mathcal{L}_n = M_n^\star,$$
$$M_n^{st}(x) = \int_x^L r_n(s)(s-x)\,ds, \quad M_B^{st} = \int_0^L s r_n(s)\,ds = M_n^\star h_n^\star.$$

$M_n^\star$ is the *partecipating modal mass* and expresses the partecipation of the different modes to the base shear, it is $\sum M_n^\star = \int_0^L m(x)\,dx$. $M_n^\star h_n^\star$ expresses the modal partecipation to base moment, $h_n^\star$ is the height where the partecipating modal mass $M_n^\star$ must be placed so that its effects on the base are the same of the static modal forces effects, or $M_n^\star$ is the resultant of s.m.f. and $h_n^\star$ is the position of this resultant.

EQ example, cantilever, 2

Starting with the definition of total mass and operating a chain of substitutions,

$$\begin{aligned} M_{\text{tot}} &= \int_0^L m(x)\,dx = \sum \int_0^L r_n(x)\,dx \\ &= \sum \int_0^L \Gamma_n m(x)\phi_n(x)\,dx = \sum \Gamma_n \int_0^L m(x)\phi_n(x)\,dx \\ &= \sum \Gamma_n \mathcal{L}_n = \sum M_n^\star, \end{aligned}$$

we have demonstrated that the sum of the partecipating modal mass is equal to the total mass.
The demonstration that $M_{B,\text{tot}} = \sum M_n^\star h_n^\star$ is similar and is left as an exercise.

EQ example, cantilever, 3

For the first 6 modes of a uniform cantilever,

n	$\mathcal{L}_n$	m_n	Γ_n	$V_{B,n}$	h_n	$M_{B,n}$
1	0.391496	0.250	1.565984	0.613076	0.726477	0.445386
2	-0.216968	0.250	-0.867872	0.188300	0.209171	0.039387
3	0.127213	0.250	0.508851	0.064732	0.127410	0.008248
4	-0.090949	0.250	-0.363796	0.033087	0.090943	0.003009
5	0.070735	0.250	0.282942	0.020014	0.070736	0.001416
6	-0.057875	0.250	-0.231498	0.013398	0.057875	0.000775
7	0.048971	0.250	0.195883	0.009593	0.048971	0.000470
8	-0.042441	0.250	-0.169765	0.007205	0.042442	0.000306

The convergence for MB is faster than for V_B, because the latter is proportional to an higher derivative of displacements.